

# EURISA: developing a European IMU for scientific and commercial applications

J.-J. Bonnefois<sup>✉</sup>, P. Cheiney<sup>✉</sup>, R. Demachy<sup>✉</sup>, S. Djeballi<sup>✉</sup>, L. Dutheil<sup>✉</sup>, L. Ferraioli<sup>✉</sup>, O. Jolly<sup>✉</sup>, R. Fredouille<sup>✉</sup>, N. Kossa<sup>✉</sup>, S. Masson<sup>✉</sup>, J.-P. Michel<sup>✉</sup>, D. Negretto<sup>✉</sup>, C. Ollivier<sup>✉</sup>, C. Roudier<sup>✉</sup>, S. Theil<sup>✉</sup>, C. Sieg<sup>✉</sup>, E. de Toldi<sup>✉</sup>, X. Weilemann<sup>✉</sup>

<sup>✉</sup>Exail / Saint-Germain-en-Laye / France

<sup>✉</sup>ETH / Zürich / Switzerland

louis.dutheil@exail.com | phone +33 1 30 08 80 32

<sup>✉</sup>Airbus Defence and Space / Toulouse / France

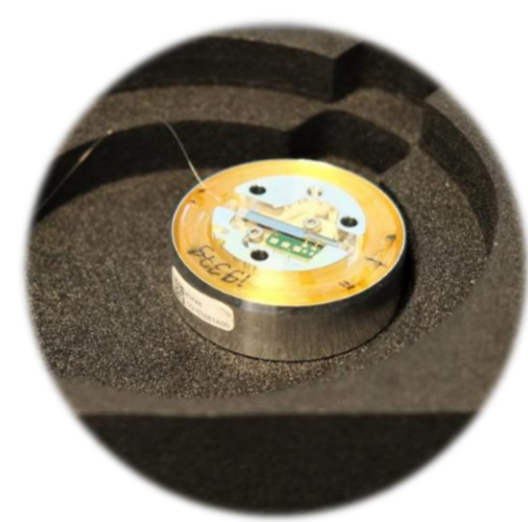
<sup>✉</sup>DLR / Bremen / Germany

## Introduction

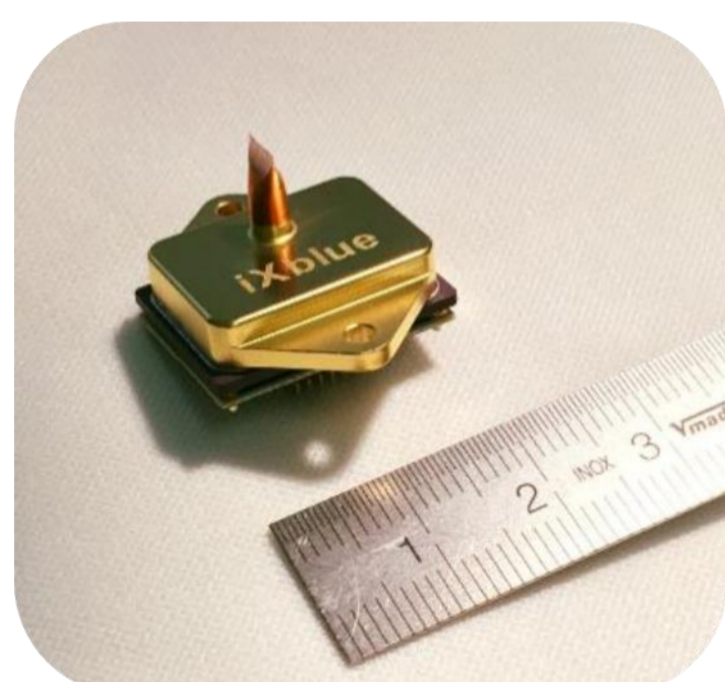
- Following the H2020 European project Pioneers on the development of the new generation of a 6-axis instrument for space seismology, EURISA aims at developing a compact, cost-efficient space IMU based on the same compact sensors.
- Propose a COTS IMU addressing of the new space applications:
  - Orbital rendez-vous
  - Carrier module
  - Moon/Mars landing
  - Mars Robotic exploration

## Sensors

- **3 FOG**  
Inherited from Exail product  
Retex on 8Mh in orbit GEO  
Magnetic shielding  
Implemented on **Astrix NS HP**  
ARW below  $< 2.5m^{\circ}/\sqrt{h}$   
No quantisation  
No other noise contributors  
Stable on all the temperature range  $[-15; 55]^{\circ}C$

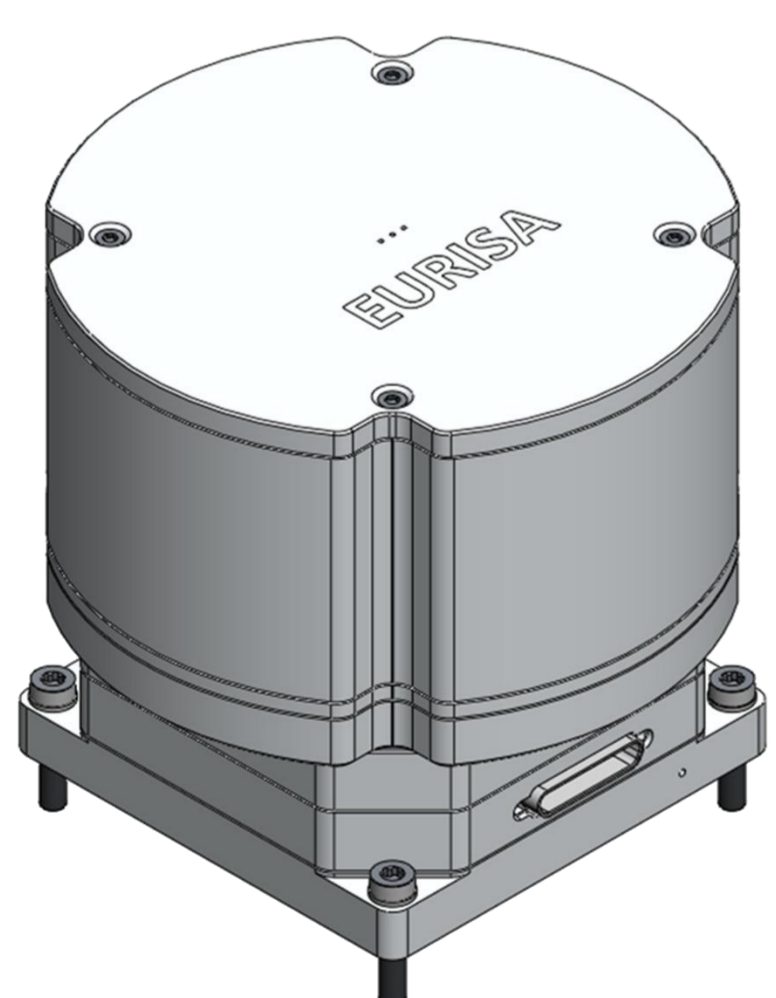


- **3 Accelerometers**  
Exail smallest ACC, already implemented on **Umix**



Digitally control sensor electronics: FPGA based  
Modelled in temperature through a thermal chamber mounted on a 3-axis rotating table → 50ppm residual scale factor

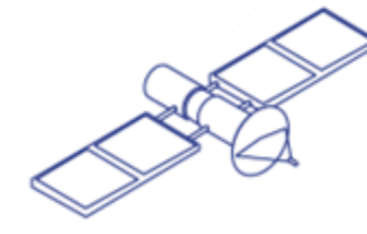
## Follow Up



- First prototype assembly late 2024
- Extensive testing at DLR Bremen  
Environmental testing  
Vibrations  
Thermal vacuum  
Electromagnetic compatibility
- Extensive testing at DLR Bremen in Spring 2025
- EM fully tested in 2025 in scope of H2020 project
- Design review in 2026
- Space qualification in 2027

## Targeted Missions

- COTS IMU for a large range of space applications.
- Each mission requirements frame the specifications



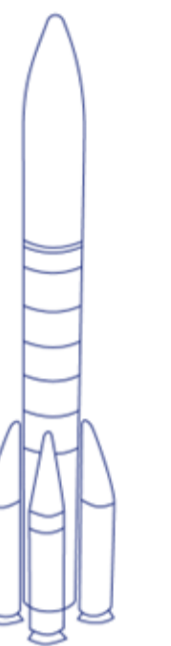
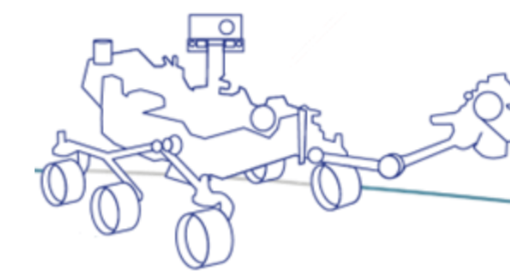
**Interplanetary cruises**  
Pointing requirements  
FOG bias (1h)  $< 0.15^{\circ}/h$   
ACC bias  $< 50\mu g$

**Entry Descent and Landing**

**Ascent Vehicle**  
Measurement range  
FOG  $\pm 100^{\circ}/s$   
ACC  $\pm 12g$   
Scale Factor  $< 150ppm$   
Full performance in harsh mechanical environment  
11gRMS

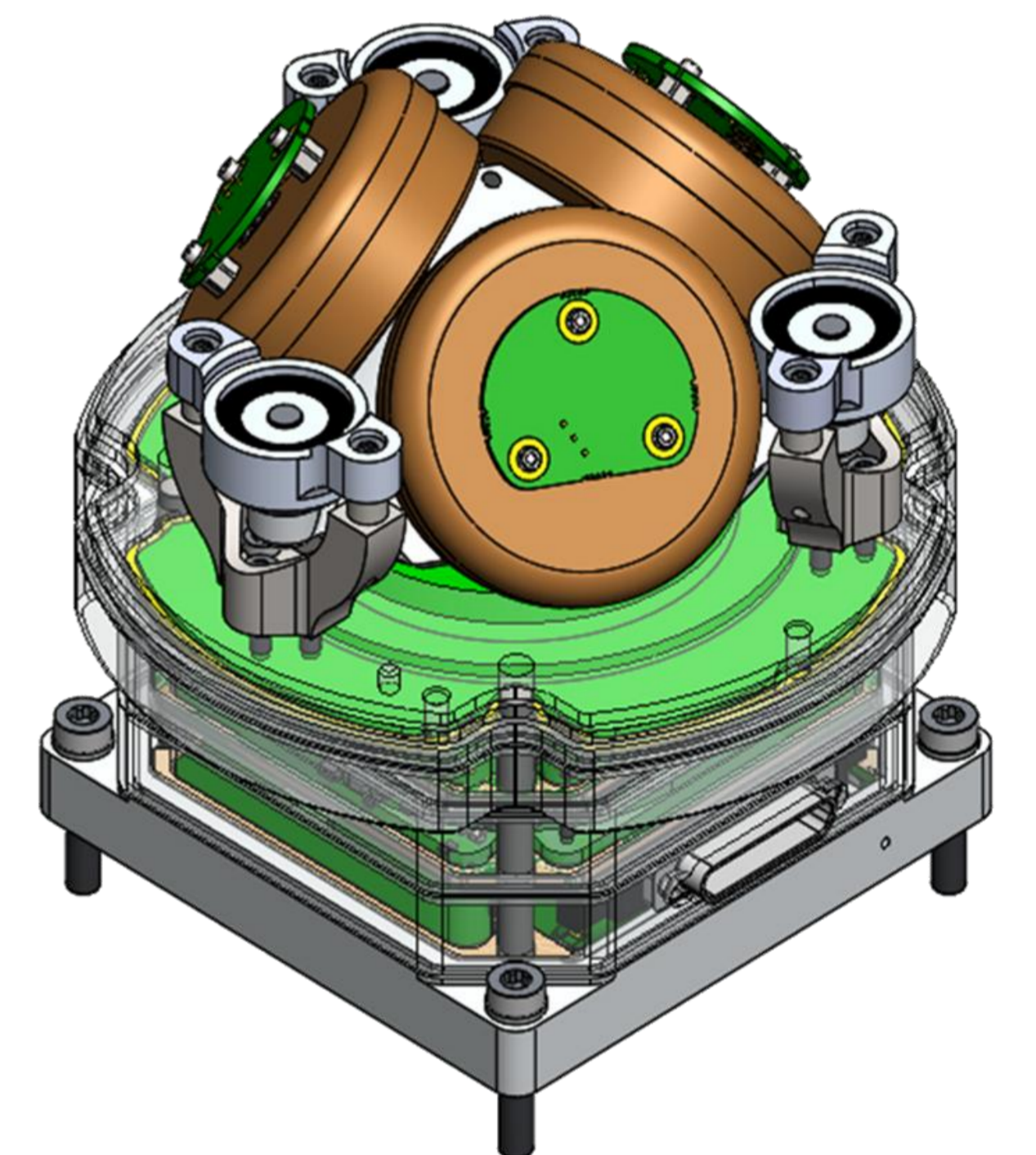


**Rover**  
Limited mechanical footprint and power efficiency



## IMU Architecture

- 3 FOG + 3 ACC mounted on a corner-cube pyramid  
Robust orthogonality and harmonisation  
Low-deformable design
- Damped pyramid  
Safe for rigorous mechanical environment
- On-board electronics  
Rad-hard electronics  
Sensors processing  
Thermal compensation (model)  
Lever arm compensation  
Coning-sculling compensation



## Consortium Partners

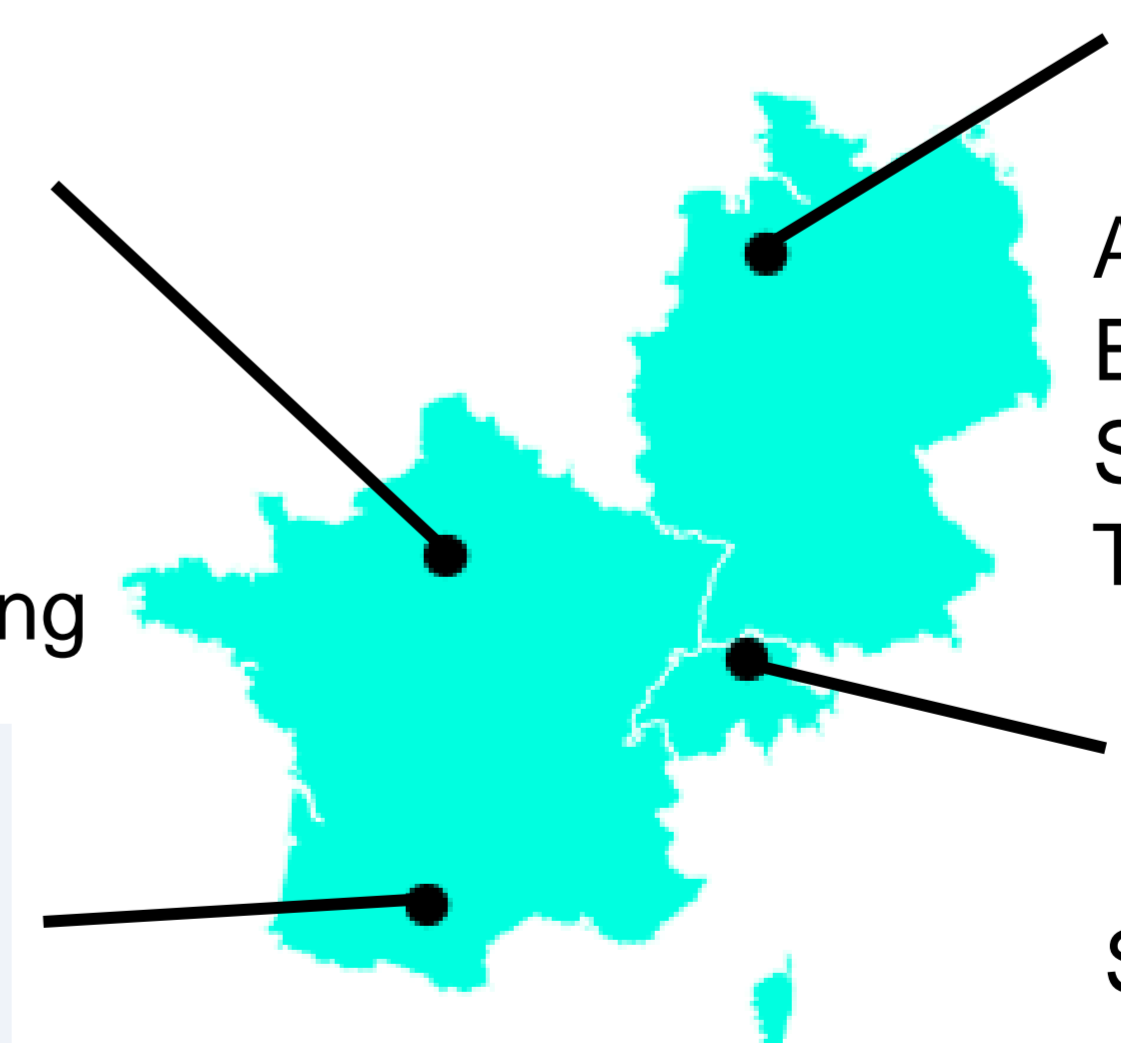
Gyroscopes  
Accelerometers  
Electronics  
Algorithms  
Assembling and testing

**AIRBUS**

Market study  
Specifications  
Expertise



Deutsches Zentrum für Luft- und Raumfahrt  
German Aerospace Center



Algorithms  
External sensor fusion  
Simulation  
Test means

**ETH zürich**

Space accelerometer control electronics



This project has received funding from the European Union's Horizon 2020 Research and Innovation Programme under grant agreement No 101004205.



Inertial Sensors and Applications  
Symposium Gyro Technology 2024

